Composite Panel Checklist Streameline Testing

TEST SPECIMEN #

<u>Please Indicate Type of Testing Preferred:</u>
Streamlined testing (FAA 8130-9 required)
Test plan requested

Company Name:		Contact Name:					
Phone:	E-Mail:						
Date Sent:	P.O. #:						
Aircraft Make:	Model:	S/N #:	Tail #:				
What area of the aircraft does this po (i.e. sidewall, cabin floor, crew seat,	•						
TEST DATA IS IN SUPPORT	OF (As Required): Field	Approval Major Repa	ir Major Alteration	Minor Repair			
Minor Alteration Organiz Technical Standard Order (T	zation Designation Author SO) Supplementa	rization (ODA) Type al Type Certificate (STC)	Certificate (TC)				
FAA form 8110-3 may only be is	sued aircraft specific for	<mark>U.S. registered or U.S. Sta</mark>	te of Design aircraft.				
Please complete checklist for each pa traceability for each panel component							
TYPES OF COMPOSITE TEST	TS AVAILABLE:						
1. Horizontal 23.853 🗌 💢	2. 12-Second Vertical 25.8	3. 60-Se	cond Vertical 25.853(a)				
4. 45-Degree Angle 25.855(d) App F Part 1 🔲	5. 60-Degree Wire	25.869(a) App F Part 1 $\ \square$				
6. Applicable Tests for the "	as installed state" for this	aircraft 🗌					
If Skandia is fabricating the test specimens, does Skandia have your permission to issue FAA 8130-9 forms on your behalf? Yes No							

REQUIREMENTS FOR SPECIMENS TO BE TESTED:

Horizontals, 12- and 60-Second Verticals less than .25": Woven specimens (fabric, carpet, etc.) three strips, 3" x 12" cut from across the roll AND three strips, 3" x 12" cut from up the roll. Non-woven specimens (leather, vinyl, plastic, foam, etc.) three strips, 3" x 12". 12- and 60-Second Verticals more than .25": Woven specimens (fabric, carpet, etc.) six strips, 3" x 12" cut from across the roll AND six strips, 3" x 12" cut from up the roll. Non-woven specimens (leather, vinyl, plastic, foam, etc.) six strips, 3" x 12". 45-Degree Angle: Three specimens, 8½" x 8½". 60-Degree Wire: Three specimens, 34" long.

Per FAA Handbook, DOT/FAA/AR-00/12 Chapter 1.0, when testing two samples per 1.6.2.4 for test specimens greater than 0.25" (6.35 mm), the recorded burn length will be from the face with the greatest burn length in accordance with section 1.6.2.9

If edge filler is used, it must be called out on the 8130-9 but should be omitted from both the material list and the composite sketch as it will be eliminated from the test specimens per 14 CFR 25.853 App F Part 1(b)(2).

As required by App F, foam shall be tested in 1/2" thicknesses. If the panel consists of two or more foams glued together, the foam specimens should be two 1/4" (three 1/16", etc.) pieces glued together per AC 25-17 Transport Airplane Cabin Interiors Crashworthiness Handbook or other accepted FAA test method.

Per FAA Fire Handbook, DOT/FAA/AR-00/12, Chapter 1, Paragraph 1.4.4.1 "For testing purpose where a part is used in several thicknesses, the minimum thickness will be tested."

Additional reference material: Advisory Circular AC 25—853-1, AC 21-25A, AC23-2 and Aircraft Materials Fire Test Handbook DOT/FAA/AR-00/12, FAA Policy Statement PS-ANM-25.853-01.

IF USING A PROCESS SPECIFICATION CALLOUT, A COPY OF THE PROCESS SPECIFICATION MUST BE INCLUDED.

Skandia, Inc. • 5000 N. Highway 251 • Davis Junction, IL 61020 • 815/393-4600 • 815/393-6878 fax • Info@SkandiaInc.com

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Composite Panel Checklist Streamline Testing

ADDITIONAL COMMENTS:

To be Tested:	Side A	Side B	Label Specimens Accordingly Both Sides	(Applies to Vertical Testing (
Composite Packing Lists	Panel Build or Invoices mu	I-Up Mater est be supplie		ecimen #
TEM NO.	ITEM TYPE		ITEM DESCRIPTION	
EXAMPLE 1	PANEL	.50" ABC C	co., P/N 012345 Aluminum Honeycomb	01-2345
EXAMPLE 2 VENEER		.030" XYZ	Corp., Walnut Burl	98765
ITEM NO.	SI	KETCH OF C	OMPOSITE BUILD-UP All Tolerances +/030	11
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CERTIFICATION FOR PART 27 AND PART 29 ROTORCRAFT (HELICOPTER), PART 23 NORMAL AND COMMUTER CATEGORY IS ALSO AVAILABLE

All work accomplished in accordance with Skandia, Inc. Work Instructions FL101-01 through FL104-01.

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